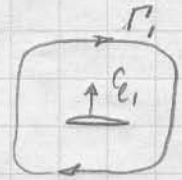


1) By Kelvin's Theorem, $\Gamma_2 = \Gamma_1$

In general, $L' = \rho V \Gamma = \frac{1}{2} \rho V_\infty^2 c c_l$

or $\Gamma = \frac{1}{2} c V_\infty c_l$ (for airfoil)



$$\therefore \Gamma_1 = \frac{1}{2} c V_\infty c_{l1} = \frac{1}{2} \cdot 1m \cdot 30m/s \cdot 0.5 = 7.5 m^2/s = \Gamma_2$$

But we also know $\Gamma_3 + \Gamma_4 = \Gamma_2$ by geometry

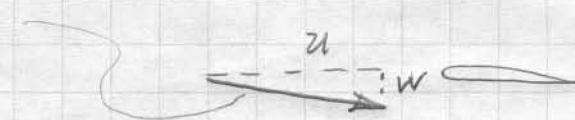
And since $\Gamma_3 = \frac{1}{2} c V_\infty c_{l2} = \frac{1}{2} \cdot 1m \cdot 30m/s \cdot 1.5 = 22.5 m^2/s$

$$\therefore \boxed{\Gamma_{\text{vortex}} = \Gamma_4 = \Gamma_2 - \Gamma_3 = -15 m^2/s} \quad (\text{counterclockwise})$$

2a) Observer sees: • horizontal $u = V_\infty = 30m/s$

• vertical $W = \frac{\Gamma_{\text{vortex}}}{2\pi d} = \frac{-15 m^2/s}{2\pi \cdot 10m} = -0.2387 m/s$

$$\vec{V}_{\text{apparent}} = u \hat{i} + w \hat{k}$$



2b) Airfoil thinks it's in a freestream $\vec{V}_{\text{apparent}}$.

Apparent Lift force is \perp to $\vec{V}_{\text{apparent}}$: $L' = \frac{1}{2} \rho |\vec{V}_{\text{apparent}}|^2 c c_l \approx \frac{1}{2} \rho V^2 c c_l$

Apparent Drag force is \parallel to $\vec{V}_{\text{apparent}} = 0$ (d'Alembert)

This Lift force is tilted back relative to motion by angle $\theta = \arctan \frac{-w}{u} = 0.456^\circ$

True drag relative to V is

$$D' = L' \sin \theta =$$

$$\text{or } \boxed{c_d = c_l \sin \theta = 0.0119}$$

